



Probing the gravitational redshift with an Earth-orbiting satellite



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ABSTRACT

We present an approach to testing the gravitational redshift effect using the RadioAstron satellite. The experiment is based on a modification of the Gravity Probe A scheme of nonrelativistic Doppler compensation and benefits from the highly eccentric orbit and ultra-stable atomic hydrogen maser frequency standard of the RadioAstron satellite. Using the presented techniques we expect to reach an accuracy of the gravitational redshift test of order 10^{-5} , a magnitude better than that of Gravity Probe A. Data processing is ongoing, our preliminary results agree with the validity of the Einstein Equivalence Principle.

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1. Introduction

Quantum theory and general relativity are the two pillars of modern physics. However, they are incompatible. Attempts to quantize gravitation in the frameworks of string theory or loop quantum gravity inevitably lead to a violation of the Einstein Equivalence Principle (EEP) and thus to a breakdown of the metric nature of gravitation [1]. Although there exist attempts to preserve the unquantized status of gravitation, they have not created a compelling case so far [2]. Tests of the EEP are therefore of primary interest to characterize any unified theory of interactions.

Much progress in the field of EEP tests was made with direct involvement and under the leadership of V. B. Braginsky, the founder of a gravitational physics school (see, e.g. [3–5]). With many coauthors of this paper proudly regarding Vladimir Borisovich as a teacher, we would like to dedicate this work to his memory.

In this paper we present an approach to probing the gravitational redshift effect, which constitutes a test of the Local Position Invariance aspect of the EEP, using the RadioAstron satellite [6]. A highly eccentric orbit and an ultra-stable on-board atomic hydrogen maser clock make RadioAstron a unique laboratory for such test. The idea of experiments of this kind is to compare the rate of time flow at different space-time points against the gravitational potential difference between them. In the simplest case when time is measured by identical clocks and the gravitational field is weak, the basic equation reads [1]:

$$\frac{\Delta T}{T} = (1 + \varepsilon) \frac{\Delta U}{c^2}, \quad (1)$$

where $\Delta T/T$ is the fractional difference of time intervals measured by the clocks, ΔU is the gravitational potential difference between them, c is the speed of light, and ε is the violation parameter to be determined. In unified theories ε is usually non-zero and depends on the clock type and element composition of the gravitational field source, while in general relativity and any other metric theory of gravitation $\varepsilon = 0$.

The concept of a satellite-based gravitational redshift experiment was developed and realized by Vessot et al. in the suborbital Gravity Probe A (GP-A) mission [7], which yielded the best such test to date: it found $\varepsilon = (0.05 \pm 1.4) \times 10^{-4}$ (for hydrogen maser clocks), with $\delta\varepsilon = 1.4 \times 10^{-4}$ constituting the accuracy of the test (1σ). A modified approach we have developed for RadioAstron allows us, in principle, to reach an accuracy of $\delta\varepsilon \sim 10^{-6}$, benefiting from a better performing hydrogen maser (H-maser) and prolonged data accumulation [8]. However, technical and operational constraints discussed below limit the achievable accuracy to $\delta\varepsilon \sim 10^{-5}$. Several competing experiments are currently at various stages of preparation or realization, with accuracy goals ranging from 4×10^{-5} to 2×10^{-6} [9–11].

The RadioAstron project is an international collaborative mission centered around the 10-m space radio telescope, with the primary goal of performing Space VLBI (Very-Long-Baseline Interferometry) observations of celestial radio sources of different nature with an extraordinary high angular resolution [6]. The RadioAstron spacecraft is on a highly eccentric orbit around the Earth, evolving due to the gravitational influence of the Moon, as well as other factors, within a broad range of the orbital parameter space (perigee altitude 1,000–80,000 km, apogee altitude 270,000–370,000 km). The gravitational redshift experiment is a part of the RadioAstron mission's Key Science Programme. The essential characteristic of the mission, making it suitable for the experiment, is the presence of the space-qualified H-maser VCH-1010 aboard the spacecraft [12].

The outline of the paper is as follows. In Section 2 we present our approach to testing the gravitational redshift effect with RadioAstron, emphasizing similarities and differences between our

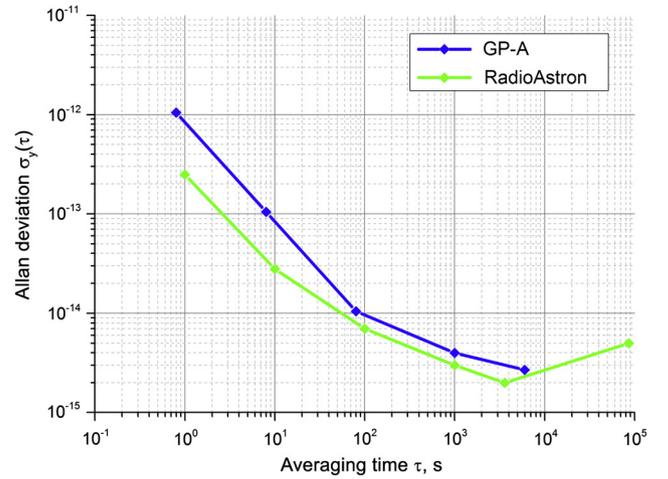


Fig. 1. Comparison of the frequency stability of the RadioAstron VCH-1010 [12] and GP-A VLG-10 [13] H-masers in terms of the Allan deviation.

Doppler compensation scheme and that of Gravity Probe A. In Section 3 we briefly discuss our data processing algorithms and describe how we treat small effects that are not cancelled by the Doppler compensation scheme. In Section 4 we give details of the measurements performed so far. We conclude with Section 5 by discussing the preliminary results and prospects for future research.

2. Outline of the RadioAstron gravitational redshift experiment

There exist two approaches to testing the gravitational redshift effect in the field of the Earth. The first one is based on measuring the total value of the gravitational redshift between a ground station and a satellite. This approach, to be pursued by the ACES mission [10] and often called the absolute gravitational redshift measurement, is feasible only with accurate clocks. The second approach, pioneered by Gravity Probe A, requires a stable clock, such as an H-maser, and is based on measuring the modulation of the redshift effect caused by the spacecraft's motion along an eccentric orbit around the Earth. We follow the second approach, benefiting from the high stability of RadioAstron's H-maser (Fig. 1) and the deep modulation of the redshift effect due to the high eccentricity of the orbit (Fig. 2). The modulation approach has an important advantage over the absolute measurement—it eliminates most systematic errors and provides for statistical averaging of the results. The ultimate goal of the experiment, in either case, is to determine the EEP violation parameter ε by comparing the experimentally measured redshift, ΔT_{grav} or Δf_{grav} , against the computed gravitational potential difference, ΔU , between the ground and spaceborne clocks using Eq. (1) or (2) (below).

In the gravitational redshift experiment with RadioAstron we detect the frequency change of RadioAstron's on-board H-maser due to gravitation by comparing it, with the help of radio links, with an H-maser at a ground station. The fractional frequency shift due to gravitation, $\Delta f_{\text{grav}}/f$, of a signal at frequency f sent from the spacecraft to a ground station, is:

$$\frac{\Delta f_{\text{grav}}}{f} = (1 + \varepsilon) \frac{\Delta U}{c^2}, \quad (2)$$

which reflects the same physics as Eq. (1). Either one of RadioAstron mission's dedicated tracking stations (TS), Pushchino (Moscow region, Russia) or Green Bank (West Virginia, USA), or a regular ground radio telescope (GRT) equipped with a 8.4 or 15 GHz receiver may be used to receive the spacecraft signal. The small gravitational frequency shift, with a maximum value of $\Delta f_{\text{grav}}/f \sim$

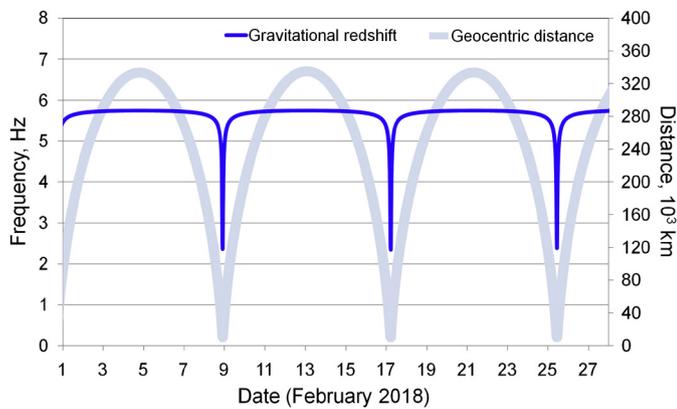


Fig. 2. Variation of the gravitational frequency shift of the 8.4 GHz downlink signal along the orbit during a low perigee epoch.

7×10^{-10} at the apogee, needs to be extracted from a number of other effects influencing the signal sent from the spacecraft to the ground station [14]:

$$\begin{aligned} \Delta f_{1w} = & \\ = f \left(-\frac{\dot{D}}{c} - \frac{v_s^2 - v_e^2}{2c^2} + \frac{(\mathbf{v}_s \cdot \mathbf{n})^2 - (\mathbf{v}_e \cdot \mathbf{n}) \cdot (\mathbf{v}_s \cdot \mathbf{n})}{c^2} \right) & \\ + \Delta f_{\text{grav}} + \Delta f_{\text{ion}} + \Delta f_{\text{trop}} + \Delta f_{\text{fine}} + \Delta f_0 + O\left(\frac{v}{c}\right)^3, & \quad (3) \end{aligned}$$

where \mathbf{v}_s and \mathbf{v}_e are the velocities of the spacecraft and the ground station (in a geocentric inertial reference frame), \dot{D} is the radial velocity of the spacecraft relative to the ground station, \mathbf{n} is a unit vector in the direction opposite to that of signal propagation, Δf_{ion} and Δf_{trop} are the ionospheric and tropospheric shifts, Δf_{fine} denotes various fine effects (phase center motion, instrumental, etc.), Δf_0 is the frequency bias between the ground and space H-masers, and “1w” stands for “1-way” (space to ground).

There are two major problems in using Eq. (3) to determine Δf_{grav} directly, at least for RadioAstron. First, the frequency bias, Δf_0 , cannot be determined after launch without making use of Eq. (2), which requires the knowledge of ε . We do not expect Δf_0 to be negligible for H-maser standards and, moreover, the well-known cavity “pulling” effect may cause it to drift over long times. We solve the bias problem by measuring only the modulation of the gravitational effect, Δf_{grav} , instead of its total value. In practice, this means having two or several observational sessions at greatly varying distances to the spacecraft. Although the measured value of the modulation of Δf_{grav} is then free from the bulk of the bias, it still includes a contribution from the bias drift. The latter, however, may be determined from a series of observations at a constant geocentric distance. The drift measured this way (Fig. 3), indeed, turns out to be non-negligible (3.6×10^{-14} /day now), so that we must take it into account.

The second problem with Eq. (3) is associated with the non-relativistic Doppler shift, $-\dot{D}/c$. Since the range rate error $\delta\dot{D}$ is ~ 2 mm/s [15], the error of the computed value of the Doppler shift is $\delta(\dot{D}/c) \sim 10^{-11}$, while 10^{-15} is required for achieving $\delta\varepsilon \sim 10^{-5}$. The first-order Doppler term, however, can be eliminated completely (for a TS), or its magnitude reduced sufficiently (for a GRT), owing to the availability of the 2-way ground-space-ground link (Fig. 4). The 2-way link signal is sent by a TS, received and phase-coherently retransmitted by the spacecraft, and finally received again by a TS and/or a GRT. The frequency shift of the 2-way link signal, for the simpler case of TS-space-TS propagation, is:

$$\Delta f_{2w} = f \left(-2\frac{\dot{D}}{c} - \frac{v_s^2 - v_e^2}{c^2} + \frac{|\mathbf{v}_s - \mathbf{v}_e|^2}{c^2} - 2\frac{\mathbf{a}_e \cdot \mathbf{n}}{c} \Delta t \right)$$

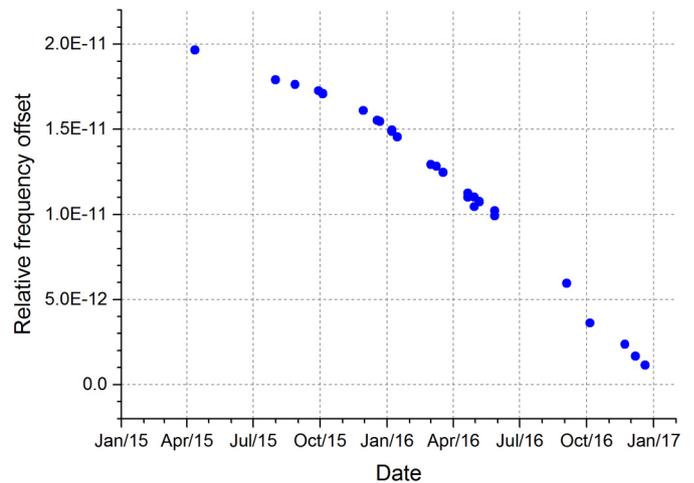


Fig. 3. Frequency drift of the RadioAstron on-board H-maser relative to the H-maser at the Green Bank TS.

$$\begin{aligned} + 2\frac{(\mathbf{v}_s \cdot \mathbf{n})^2 - (\mathbf{v}_e \cdot \mathbf{n}) \cdot (\mathbf{v}_s \cdot \mathbf{n})}{c^2} & \\ + 2\Delta f_{\text{trop}} + 2\Delta f_{\text{ion}} + O(v/c)^3, & \quad (4) \end{aligned}$$

where \mathbf{a}_e is the ground station acceleration and Δt is the signal's light travel time [14]. (A physically similar but computationally more complex equation holds for the case of the 2-way link signal received by a nearby GRT.) Combining the 1-way (3) and 2-way (4) frequency measurements, we obtain:

$$\begin{aligned} \Delta f_{1w} - \frac{\Delta f_{2w}}{2} = \Delta f_{\text{grav}} + f \left(-\frac{|\mathbf{v}_s - \mathbf{v}_e|^2}{2c^2} + \frac{\mathbf{a}_e \cdot \mathbf{n}}{c} \Delta t \right) & \\ + \Delta f_0 + \Delta f_{\text{ion}}^{(\text{res})} + \Delta f_{\text{fine}} + O(v/c)^4, & \quad (5) \end{aligned}$$

where $\Delta f_{\text{ion}}^{(\text{res})}$ is the residual ionospheric shift (fully suppressed only for equal up- and downlink frequencies) and Δf_{fine} denotes several “fine” effects, such as those due to the relativistic kinematic terms of order $(v/c)^3$:

$$\begin{aligned} \frac{\Delta f^{(3)}}{f} = \frac{\mathbf{n} \cdot (\mathbf{v}_e - \mathbf{v}_s)}{c^3} \left(\Delta U - \frac{|\mathbf{v}_s - \mathbf{v}_e|^2}{2} + (\mathbf{a}_e \cdot \mathbf{n}) c \Delta t \right) & \\ + \frac{D}{c^3} (-\mathbf{v}_s \cdot \mathbf{a}_e - (\mathbf{j}_e \cdot \mathbf{n}) c \Delta t + 2\mathbf{v}_e \cdot \mathbf{a}_e + \mathbf{v}_e \cdot \nabla U_e), & \quad (6) \end{aligned}$$

where $\mathbf{j}_e = \dot{\mathbf{a}}_e$ is the ground station jerk and ∇U_e is the gradient of the gravitational potential at the ground station location (see Section 3 for other fine effects). It is important to note that Eq. (5) is free from the nonrelativistic Doppler and tropospheric effects but retains the contribution of gravitation. The idea of the compensation scheme based on Eq. (5) was first realized in the GP-A mission, and the necessity of taking into account third-order kinematic effects was noted in [16,17]. For RadioAstron, however, this scheme is not directly applicable because 1- and 2-way links cannot be operated simultaneously (Fig. 4). Nevertheless, two options for realizing the compensation scheme of Eq. (5) with RadioAstron have been devised.

The first option requires interleaving the 1-way “H-maser” (Fig. 4a) and 2-way “Coherent” (Fig. 4b) operation modes. The data recorded by GRTs (and the TS) contain only one kind of signal at any given time. However, if the switching cycle is short enough (~ 4 min at 8.4 GHz) we can interpolate the phases into the gaps with a corresponding frequency error of $\Delta f/f \sim 4 \times 10^{-15}$. Thus we obtain simultaneous frequency measurements of both kinds and can apply the compensation scheme of Eq. (5) to them directly. The approach based on interleaved measurements does not

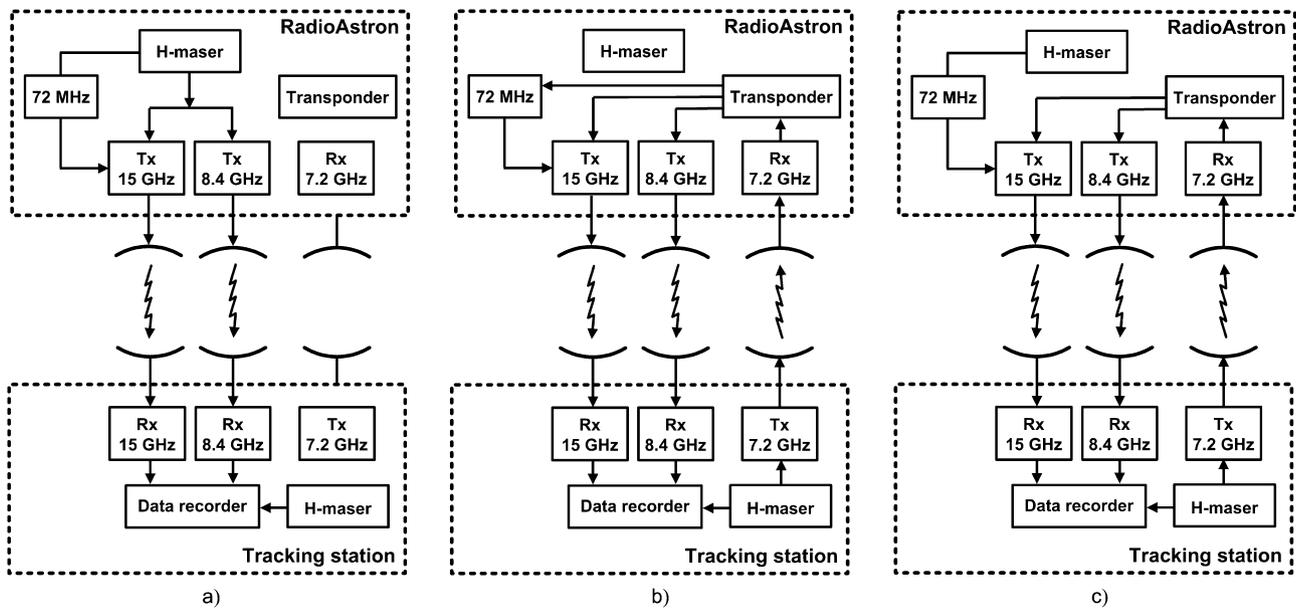


Fig. 4. Operation modes of the RadioAstron radio links: a) “H-Maser” (1-way); b) “Coherent” (2-way); c) “Semi-Coherent” (2-way).

rely on any features of the signal spectrum and may be realized with telescopes equipped either with 8.4 or 15 GHz receivers.

The second option for the Doppler compensation involves recording the 15 GHz data downlink signal in the “Semi-Coherent” mode of synchronization of the on-board scientific and radio equipment [8], which is a kind of half-way between the 1-way “H-maser” and 2-way “Coherent” modes. In this mode the 7.2 GHz uplink tone, the 8.4 GHz downlink tone and the carrier of the 15 GHz data downlink are phase-locked to the ground H-maser signal, while the modulation frequency (72 MHz) of the data downlink is phase-locked to the on-board H-maser signal (Fig. 4c). This approach relies on the broadband (~ 1 GHz) nature of the 15 GHz signal modulated using quadrature phase-shift keying (QPSK) and the possibility of turning its spectrum into a comb-like form by transmitting a predefined periodic data sequence (Fig. 5). As we have shown in [8], different subtones of the resulting spectrum act like separate links of the GP-A scheme and can be arranged in software postprocessing into a combination similar to that of Eq. (5), which is free from the 1st-order Doppler and tropospheric effects (the ionospheric term persists).

Despite some advantages of the second option from algorithmic and operational points of view, we give preference to the interleaved measurements approach as it provides for a larger number of participating GRTs due to the larger ground footprint of the on-board antenna at 8.4 GHz and wider availability of 8.4 GHz receivers at GRTs.

3. Data processing and fine effects

The primary data for the experiment are the spacecraft signals at 8.4 and/or 15 GHz received and recorded at a ground station. The majority of radio astronomy and geodetic radio telescopes are equipped with H-maser standards and 8.4 GHz receivers, enabling them to take part in the experiment. Recording of the spacecraft signal is performed in the ground H-maser timescale using standard VLBI back-end instrumentation. Initial data processing is based on the algorithms developed originally for PRIDE (Planetary Radio Interferometry and Doppler Experiment) [18] for recovering the phase of the received signal. Details of the algorithm and software modifications required to process interleaved data will be given in an upcoming publication [19]. Here we briefly describe the approaches for correcting the recovered signal phases

for a number of fine effects contributing to the right-hand side of Eq. (5):

- second- and third-order relativistic kinematic effects: computed from the orbital data (velocity determination accuracy of $\delta v \sim 2$ mm/s [15] is sufficient);
- gravitational potential difference between the spacecraft and the ground station: computed from the orbital data using the Earth gravitational potential model [20] (the position error of ~ 200 m provided by radio ranging [15] is sufficient for distances $\gtrsim 40,000$ km, laser ranging required otherwise);
- residual ionospheric frequency shift: computed from 2-frequency measurements (8.4 and 15 GHz), ionospheric total electron content (TEC) maps [21] and mapping functions [22], onsite GNSS receiver measurements;
- frequency shift due to the tidal gravitational field of the Sun and Moon [17]: computed from the planetary and lunar ephemerides (JPL DE430);
- phase center motion of the on-board and tracking station antennas: computed from the orbital and housekeeping data [23];
- temperature dependence of the on-board H-maser: computed from the H-maser sensitivity determined during ground tests and housekeeping data;
- magnetic field dependence of the on-board H-maser frequency: computed from the H-maser sensitivity determined during ground tests, the magnetic field model [24] and the orbital data;
- ground station motion due to solid Earth tides: computed from Earth models [25].

After the gravitational frequency shift has been measured experimentally in a series of observations at various distances on a single orbit, we fit it against the gravitational potential difference according to Eq. (2), thus obtaining a single measurement of ε . The accuracy of the result of 2–3 years of planned data accumulation depends on the number of experiments performed and their parameters. Based on the experiment error budget [26] and taking into account the observations performed so far and those planned we expect the accuracy of the test to reach $\delta\varepsilon \sim 10^{-5}$.

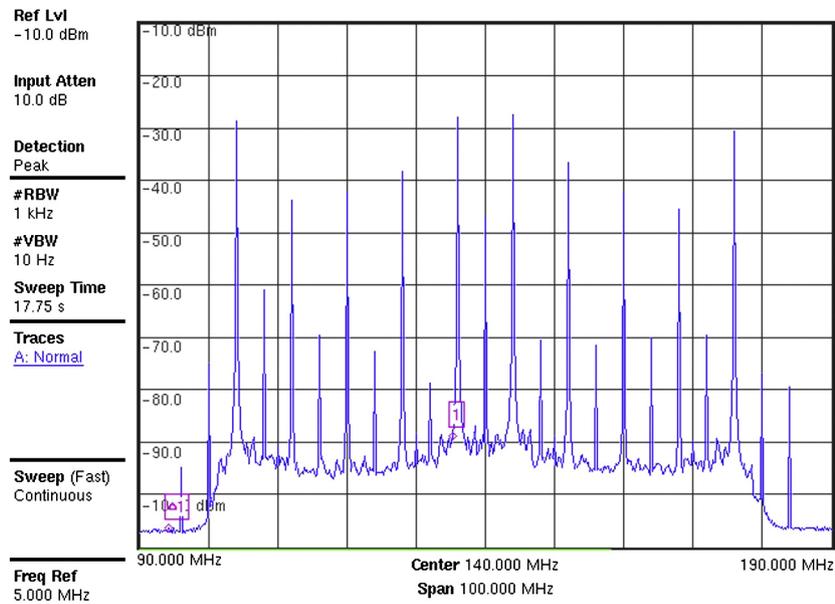


Fig. 5. Spectrum analyzer screenshot, showing the signal spectrum of the 15 GHz data downlink in the “Test-2” mode of the on-board formatter.

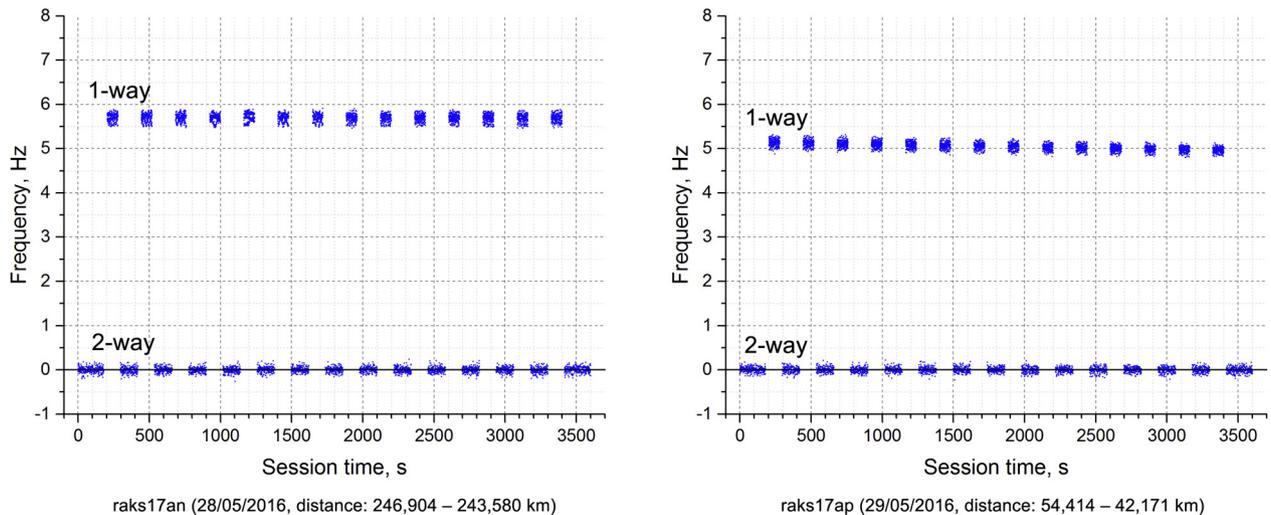


Fig. 6. Residual frequencies of the 1- and 2-way 8.4 GHz signals measured with the Onsala 20-m telescope. The 1-way frequency residuals are not corrected for the gravitational redshift. This makes the variation of the gravitational frequency shift between the two sessions clearly visible (varying from 5.69 Hz to 4.96 Hz).

4. Measurements

The observations for the experiment are limited both by the technical constraints of the RadioAstron satellite and competition for observational time with other science projects of the RadioAstron mission. The technical constraints are:

- the spacecraft’s attitude limitations with respect to the Earth, Sun and Moon;
- the requirement that the spacecraft must be visible by the particular ground antenna;
- the requirement that the ground antenna must be within the on-board antenna’s ground footprint.

A total of 18 experiments have been performed so far with the RadioAstron mission’s Pushchino and Green Bank tracking stations supported by several European VLBI Network telescopes (Effelsberg, Onsala, Svetloe, Wettzell, Yebes, Zelenchukskaya), the Robert C. Byrd Green Bank Telescope, and several Very Long Baseline Array antennas. Extensive tests have been performed with the ra-

dio telescopes and satellite laser ranging facilities of the Hartbeesthoek Radio Astronomy Observatory and AuScope VLBI Project’s Yarragadee Observatory, which are to join the observations. All experiments were performed in the interleaved measurements mode (Fig. 6) and consisted of up to 4 observations, each ~ 1 hr long, distributed along the orbit over ~ 20 –50 hr. Most observations were accompanied by satellite laser ranging to guarantee an orbit determination accuracy at the cm-level [27]. The evaluation of our preliminary experimental results, which are consistent with $\varepsilon = 0$, will be published elsewhere.

5. Conclusions

The RadioAstron satellite, with its highly eccentric orbit and on-board H-maser frequency standard, is a unique space-borne laboratory for probing the gravitational redshift effect, which constitutes a Local Position Invariance test of the Einstein Equivalence Principle. We have developed, and are implementing, a strategy for using the RadioAstron satellite to measure the gravitational redshift which takes into account the limitations of the spacecraft. We

should be able to measure the redshift to an accuracy of $\sim 10^{-5}$, which is an order of magnitude better than the current best result of Gravity Probe A. Several measurements have already been obtained, and the data are being analyzed. Our preliminary results agree with the validity of the EEP. Some of the techniques we have developed, e.g. the particular realization of the Doppler compensation scheme, could be used for future space missions to test fundamental physics.

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